

IN THE SPECIFICATION

Please delete Paragraph [0002] and replace it with the following:

[0002] At least some known turbine engines include a compressor for compressing air which is suitably mixed with a fuel and channeled to a combustor wherein the mixture is ignited within a combustion chamber for generating hot combustion gases. A portion of the compressor discharge air is directed to a cavity surrounding the combustor for cooling the combustor liner. An operating temperature of the compressor discharge air is at least partially determinative of a rating and power capability of the engine. Accordingly, within at least some known engines the compressor ~~discharge~~ discharge air temperature is monitored with a temperature probe.

Please delete Paragraph [0012] and replace it with the following:

[0012] Figure 2 is a cross-sectional view of an exemplary gas turbine engine combustor, such as combustor 16, that may be used with engine 10 (shown in Figure 1) and includes an instrument probe mounting assembly for removably coupling an instrument probe to the engine. An annular combustion chamber 90 is defined within an annular combustor outer casing 92. Combustor outer casing 92 is spaced ~~radially~~ radially outward from chamber 90 and defines an outer surface of an combustor flow channel 94 for compressor discharge air to pass therethrough for cooling purposes. Combustion chamber 90 includes an annular combustor outer liner 96 and an annular combustor inner liner 98, and extends axially downstream 99 to a high pressure turbine nozzle 95. An upstream end 97 of combustion chamber 90 includes an annular dome 100 that includes a plurality of air entry holes to admit compressor discharge air. A cross-sectional area of combustion chamber 90 diminishes in the downstream direction 99 to correspond to a cross sectional area of high pressure turbine nozzle 95.